INTERNATIONAL STANDARD

ISO 12108

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Metallic materials — Fatigue testing — Fatigue crack growth method

Matériaux métalliques — Essais de fatigue — Méthode d'essai de propagation de fissure en fatigue

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Foreword

ISO (the International Organization for Standardization) is a worldwide federation of national standards bodies (ISO member bodies). The work of preparing International Standards is normally carried out through ISO technical committees. Each member body interested in a subject for which a technical committee has been established has the right to be represented on that committee. International organizations, governmental and non-governmental, in liaison with ISO, also take part in the work. ISO collaborates closely with the International Electrotechnical Commission (IEC) on all matters of electrotechnical standardization.

International Standards are drafted in accordance with the rules given in the ISO/IEC Directives, Part 3.

Draft International Standards adopted by the technical committees are circulated to the member bodies for voting. Publication as an International Standard requires approval by at least 75 % of the member bodies casting a vote.

Attention is drawn to the possibility that some of the elements of this International Standard may be the subject of patent rights. ISO shall not be held responsible for identifying any or all such patent rights.

International Standard ISO 12108 was prepared by Technical Committee ISO/TC 164, *Mechanical testing of metals*, Subcommittee SC 5, *Fatigue testing*.

Introduction

This International Standard is intended to provide specifications for generation of fatigue crack growth rate data. Test results are expressed in terms of the fatigue crack growth rate as a function of crack-tip stress intensity factor range, ΔK, as defined by the theory of linear elastic fracture mechanics [1]-[6]. Expressed in these terms the results characterize a material's resistance to subcritical crack extension under cyclic force test conditions. This resistance is independent of specimen planar geometry and thickness, within the limitations specified in clause 6. All values are given in SI units [7].

This International Standard describes a method of subjecting a precracked notched specimen to a cyclic force. The crack length, a, is measured as a function of the number of elapsed force cycles, N. From the collected crack length and corresponding force cycles relationship the fatigue crack growth rate, da/dN, is determined and is expressed as a function of stress intensity factor range, ΔK.

Materials that can be tested by this method are limited by size, thickness and strength only to the extent that the material must remain predominantly in an elastic condition during testing and that buckling is precluded.

Specimen size may vary over a wide range. Proportional planar dimensions for six standard configurations are presented. The choice of a particular specimen configuration may be dictated by the actual component geometry, compression test conditions or suitability for a particular test environment.

Specimen size is a variable that is subjective to the test material's 0,2 % proof strength and the maximum stress intensity factor applied during test. Specimen thickness may vary independent of the planar size, within defined limits, so long as large-scale yielding is precluded and out-of-plane distortion or buckling is not encountered. Any alternate specimen configuration other than those included in this International Standard may be used, provided there exists an established stress intensity factor calibration expression, i.e. stress intensity factor function, $g\left(a/W\right)$ [9]-[11].

Residual stresses [12], [13], crack closure [14], [15], specimen thickness, cyclic waveform, frequency and environment, including temperature, may markedly affect the fatigue crack growth data but are in no way reflected in the computation of ΔK, and so should be recognized in the interpretation of the test results and be included as part of the test report. All other demarcations from this method should be noted as exceptions to this practice in the final report.

For crack growth rates above 10^{-5} mm/cycle the typical scatter in test results generated in a single laboratory for a given ΔK can be in the order of a factor of two [16]. For crack growth rates below 10^{-5} mm/cycle, the scatter in the da/dN calculation may increase to a factor of 5 or more. To assure the correct description of the material's da/dN versus ΔK behaviour, a replicate test conducted with the same test parameters is highly recommended.

Service conditions may exist where varying ΔK under conditions of constant K_{max} or K_{mean} control [17] may be more representative than data generated under conditions of constant force ratio; however, these alternate test procedures are beyond the scope of this International Standard.

Metallic materials — Fatigue testing — Fatigue crack growth method

1 Scope

This International Standard describes tests for determining the fatigue crack growth rate from the threshold stress-intensity factor range, ΔK_{th}, to the onset of unstable crack extension as the maximum stress intensity factor approaches K_{max} controlled instability, as determined in accordance with ISO 12737 [8].

This International Standard is primarily intended for use in evaluating isotropic metallic materials under predominantly linear-elastic stress conditions and with force applied only perpendicular to the crack plane (mode I stress condition), and with a constant stress ratio, R.

2 Normative reference

The following normative document contains provisions which, through reference in this text, constitute provisions of this International Standard. For dated references, subsequent amendments to, or revisions of, this publication do not apply. However, parties to agreements based on this International Standard are encouraged to investigate the possibility of applying the most recent edition of the normative document indicated below. For undated references, the latest edition of the normative document referred to applies. Members of ISO and IEC maintain registers of currently valid International Standards.

ISO 4965:1979, *Axial load fatigue testing machines — Dynamic force calibration — Strain gauge technique*

3 Terms and definitions

For the purposes of this International Standard, the following terms and definitions apply.

3.1

crack length

a

linear measure of a principal planar dimension of a crack from a reference plane to the crack tip, also called crack size

3.2

cycle

N

smallest segment of a force-time or stress-time function which is repeated periodically

NOTE The terms fatigue cycle, force cycle and stress cycle are used interchangeably. The letter N is used to represent the number of elapsed force cycles.

3.3

fatigue crack growth rate

da/dN

extension in crack length per force cycle

3.4
maximum force
F_{max}
force having the highest algebraic value in the cycle; a tensile force being positive and a compressive force being negative

3.5
minimum force
F_{min}
force having the lowest algebraic value in the cycle; a tensile force being positive and a compressive force being negative

3.6
force range
ΔF
the algebraic difference between the maximum and minimum forces in a cycle

$$\Delta F = F_{\text{max}} - F_{\text{min}}$$

3.7
force ratio
R
algebraic ratio of the minimum force or stress to maximum force or stress in a cycle

$$R = F_{\text{min}} / F_{\text{max}}$$

NOTE It is also called stress ratio.

3.8
stress intensity factor
K
magnitude of the ideal crack tip stress field for the opening mode force application to a crack in a homogeneous, linear-elastically stressed body where opening mode of a crack corresponds to the force being applied to the body perpendicular to the crack faces only (mode I stress condition)

3.9
maximum stress intensity factor
K_{max}
highest algebraic value of the stress intensity factor in a cycle, corresponding to F_{max}

3.10
minimum stress intensity factor
K_{min}
lowest algebraic value of the stress intensity factor in a cycle, corresponding to F_{min}

NOTE This definition remains the same, regardless of the minimum force being tensile or compressive. For a negative force ratio ($R < 0$) there is an alternate, commonly used definition for the minimum stress intensity factor, $K_{\text{min}} = 0$. See 3.11.

3.11
stress intensity factor range
ΔK
algebraic difference between the maximum and minimum stress intensity factors in a cycle

$$\Delta K = K_{\text{max}} - K_{\text{min}}$$

NOTE The force variables ΔK, R and K_{max} are related as follows: $\Delta K = (1 - R)\, K_{\text{max}}$. For a negative force ratio ($R < 0$) there is an alternate, commonly used definition for the stress intensity factor range, $\Delta K = K_{\text{max}}$. See 3.10 and 10.6.

3.12
fatigue crack growth threshold
ΔK_{th}
asymptotic value of ΔK for which da/dN approaches zero

NOTE For most materials the threshold is defined as the stress intensity factor range corresponding to 10^{-8} mm/cycle. When reporting ΔK_{th}, the corresponding lowest decade of da/dN data used in its determination should also be included.

3.13
normalized K-gradient
$C = (1/K)\, \mathrm{d}K/\mathrm{d}a$
fractional rate of change of K with increased crack length, a

$$C = 1/K\,(\mathrm{d}K/\mathrm{d}a) = 1/K_{max}\,(\mathrm{d}K_{max}/\mathrm{d}a) = 1/K_{min}\,(\mathrm{d}K_{min}/\mathrm{d}a) = 1/\Delta K\,(\mathrm{d}\Delta K/\mathrm{d}a)$$

3.14
K-decreasing test
test in which the value of the normalized K-gradient, C, is negative

NOTE A K-decreasing test is conducted by reducing the stress intensity factor either by continuously shedding or by a series of steps, as the crack grows.

3.15
K-increasing test
test in which the value of C is positive

NOTE For standard specimens, a constant force amplitude results in a K-increasing test where the value of C is positive and increasing.

3.16
stress intensity factor function
$g\,(a/W)$
mathematical expression, based on experimental, numerical or analytical results, that relates the stress intensity factor to force and crack length for a specific specimen configuration

4 Symbols and abbreviations

4.1 Symbols

See Table 1.

Table 1 — Symbols and their designations

Symbol	Designation	Unit
Loading		
C	Normalized K-gradient	mm^{-1}
E	Tensile modulus of elasticity	MPa
F	Force	kN
F_{max}	Maximum force	kN
F_{min}	Minimum force	kN
ΔF	Force range	kN
K	Stress intensity factor	$MPa{\cdot}m^{1/2}$
K_{max}	Maximum stress intensity factor	$MPa{\cdot}m^{1/2}$
K_{min}	Minimum stress intensity factor	$MPa{\cdot}m^{1/2}$
ΔK	Stress intensity factor range	$MPa{\cdot}m^{1/2}$
ΔK_i	Initial stress intensity factor range	$MPa{\cdot}m^{1/2}$

Table 1 — Symbols and their designations (*continued*)

Symbol	Designation	Unit
ΔK_{th}	Threshold stress intensity factor range	$MPa \cdot m^{1/2}$
N	Number of cycles	1
R	Force ratio or stress ratio	1
R_m	Ultimate tensile strength at the test temperature	MPa
$R_{p0,2}$	0,2 % proof strength at the test temperature	MPa
Geometry		
a	Crack length or size measured from the reference plane to the crack tip	mm
a_{cor}	Crack front curvature correction length	mm
a_{fat}	Fatigue crack length measured from the notch root	mm
a_n	Machined notch length	mm
a_p	Precrack length	mm
B	Specimen thickness	mm
D	Hole diameter for CT, SENT or CCT specimen, loading tup diameter for bend specimens	mm
$g\,(a/W)$	Stress intensity factor function	1
h	Notch height	mm
W	Specimen width, distance from reference plane to edge of specimen	mm
$(W-a)$	minimum uncracked ligament	mm
Crack growth		
$\mathrm{d}a/\mathrm{d}N$	Fatigue crack growth rate	mm/cycle
Δa	Change in crack length, crack extension	mm

4.2 Abbreviations for specimen identification

CT Compact tension

CCT Centre cracked tension

SENT Single edge notch tension

SENB3 Three-point single edge notch bend

SENB4 Four-point single edge notch bend

SENB8 Eight-point single edge notch bend

5 Apparatus

5.1 Testing machine

5.1.1 General

The testing machine shall have smooth start-up and a backlash-free force train if passing through zero force. See ISO 4965. Cycle to cycle variation of the peak force during precracking shall be less than $\pm$ 5 % and shall be held to within $\pm$ 2 % of the desired peak force during the test. ΔF shall also be maintained to within $\pm$ 2 % of the desired range during test. A practical overview of test machines and instrumentation is available [33], [34].

5.1.2 Testing machine alignment

It is important that adequate attention be given to alignment of the testing machine and during machining and installation of the grips in the testing machine.

For tension-compression testing, the length of the force train should be as short and stiff as practical. Non-rotating joints should be used to minimize off-axis motion.

Asymmetry of the crack front is an indication of misalignment; a strain gauged specimen similar to the test article under investigation can be used in aligning the force train and to minimize nonsymmetrical stress distribution and/or bending strain to less than 5 %.

5.1.3 Force measuring system

Accuracy of the force measuring system shall be verified periodically in the testing machine. The calibration for the force transducer shall be traceable to a national organization of metrology. The force measuring system shall be designed for tension and compression fatigue testing and possess great axial and lateral rigidity. The indicated force, as recorded as the output from the computer in an automated system or from the final output recording device in a noncomputer system, shall be within the permissible variation from the actual force. The force transducer's capacity shall be sufficient to cover the range of force measured during a test. Errors greater than 1 % of the difference between minimum and maximum measured test force are not acceptable.

The force measuring system shall be temperature compensated, not have zero drift greater than 0,002 % of full scale, nor have a sensitivity variation greater than 0,002 % of full scale over a 1 $^{\circ}$C change. During elevated and cryogenic temperature testing, suitable thermal shielding/compensation shall be provided to the force measuring system so it is maintained within its compensation range.

5.2 Cycle-counter

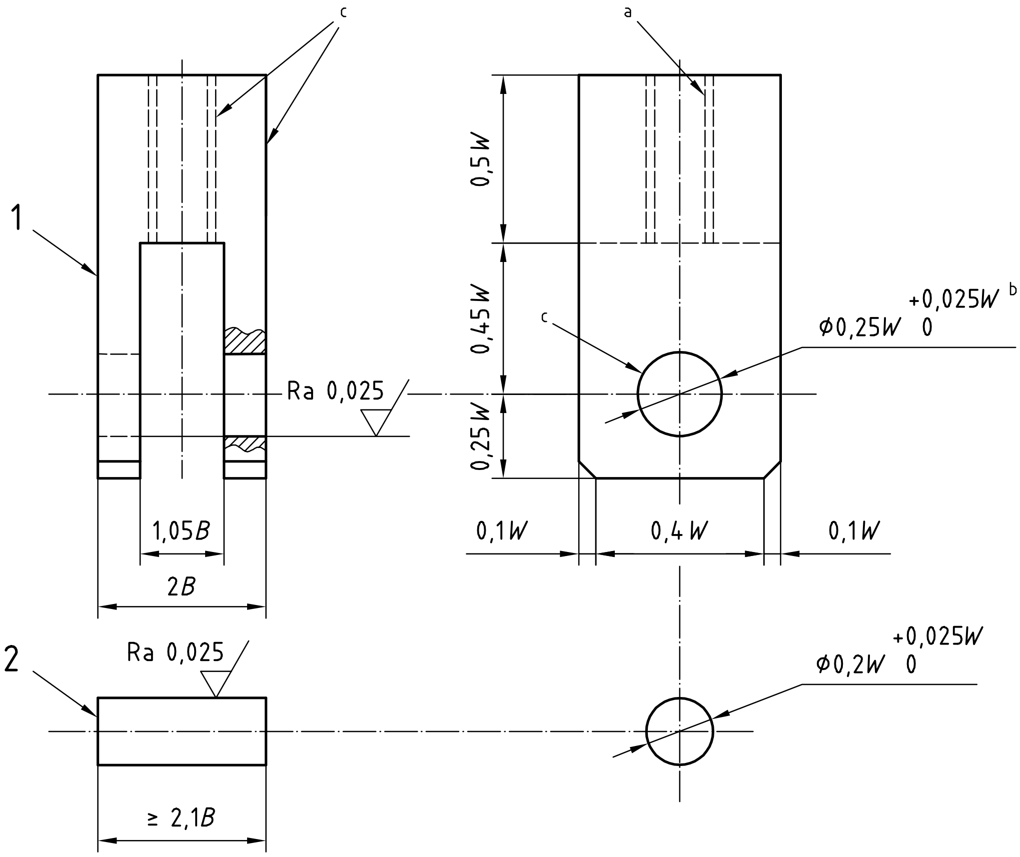
An accurate digital device is required to count elapsed force cycles. A timer is to be used only as a verification check on the accuracy of the counter. It is preferred that individual force cycles be counted. However, when the crack velocity is below 10^{-5} mm/cycle counting in increments of ten cycles is acceptable.

5.3 Grips and fixtures for CT specimens

Force is applied to a CT specimen through pinned joints. Choice of this specimen and gripping arrangement necessitates tension-tension test conditions only. Figure 1 shows the clevis and mating pin assembly used at both the top and bottom of a CT specimen to apply the force perpendicular to the machined starter notch and crack plane. Suggested dimensions are expressed as a proportion of specimen width, W, or thickness, B, since these dimensions can vary independently within the limits specified in clause 6. The pin holes have a generous clearance over the pin diameter, 0,2W minimum, to minimize resistance to specimen and pin in-plane rotation which has been shown to cause nonlinearity in the force versus displacement response [35]. A surface finish range of 0,8 µm to 1,6 µm is suggested for grip surfaces. With this grip and pin arrangement, materials with low proof strength may sustain plastic deformation at the specimen pin hole; similarly, when testing high strength materials and/or when the clevis displacement exceeds 1,05B, a stiffer force pin, i.e. a diameter greater than 0,225W, may be required. As an alternative approach to circumvent plastic deformation, a flat bottom clevis hole may be used along with a pin diameter equalling 0,24W. Any heat treatable steel thermally processed to a 0,2 % proof strength of 1 000 MPa used in fabricating the clevises will usually provide adequate strength and resistance to fretting, galling and fatigue.

In addition to the generous pin hole clearance, the mating surfaces shall be prepared to minimize friction which could invalidate the provided K-calibration expression. The use of high viscosity lubricants and greases has been shown to cause hysteresis in the force versus displacement response and is not recommended if compliance measurements are required.

Surface roughness values in micrometres



Key

1 Loading rod

2 Pin

NOTE For high strength materials or large pin displacements the pin may be stiffened by increasing the diameter to 0,24W along with using D-shaped flat bottom holes.

[a] Loading rod thread.

[b] Through diameter.

[c] These surfaces are perpendicular and parallel as applicable to within 0,05W.

Figure 1 — Clevis and pin assembly for gripping a CT specimen

5.4 Grips and fixtures for CCT/SENT specimens

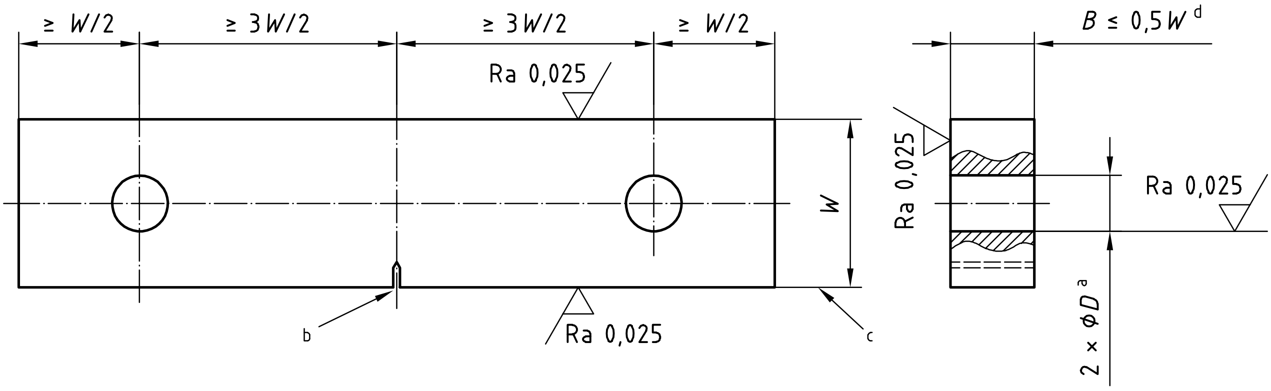
5.4.1 General

Force can be applied to CCT and SENT specimens through pinned joints and/or through frictional clamping grips. Gripping for the CCT and SENT specimens depends on specimen width and whether the test condition is to be tension-tension or tension-compression. The minimum CCT specimen gauge length varies with gripping arrangement and shall provide a uniform stress distribution in the gauge length during the test.

Equation (6) is applicable only for a single pinned end SENT specimen, as shown in Figure 2. The SENT pinned end specimen (Figure 2) is appropriate for tension-tension test conditions only.

Equation (7) is applicable for a SENT specimen with clamped ends and is appropriate for both tension and compression force conditions. For the clamped end SENT specimen, the grips must be sufficiently stiff to circumvent any rotation of the specimen ends or any lateral movement of the crack plane; the presence of either condition introduces errors into the stress intensity factor calculation [29].

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,005W.

NOTE 2 The surfaces are parallel and perpendicular to within $\pm$ 0,002W.

NOTE 3 The crack length is measured from the reference loading plane containing the starter V-notch.

NOTE 4 Specimen recommended for notch root tension at a force ratio $R > 0$ only.

a $D = W/3$.

b See Figure 12 for notch detail.

c Reference plane.

d The recommended thickness.

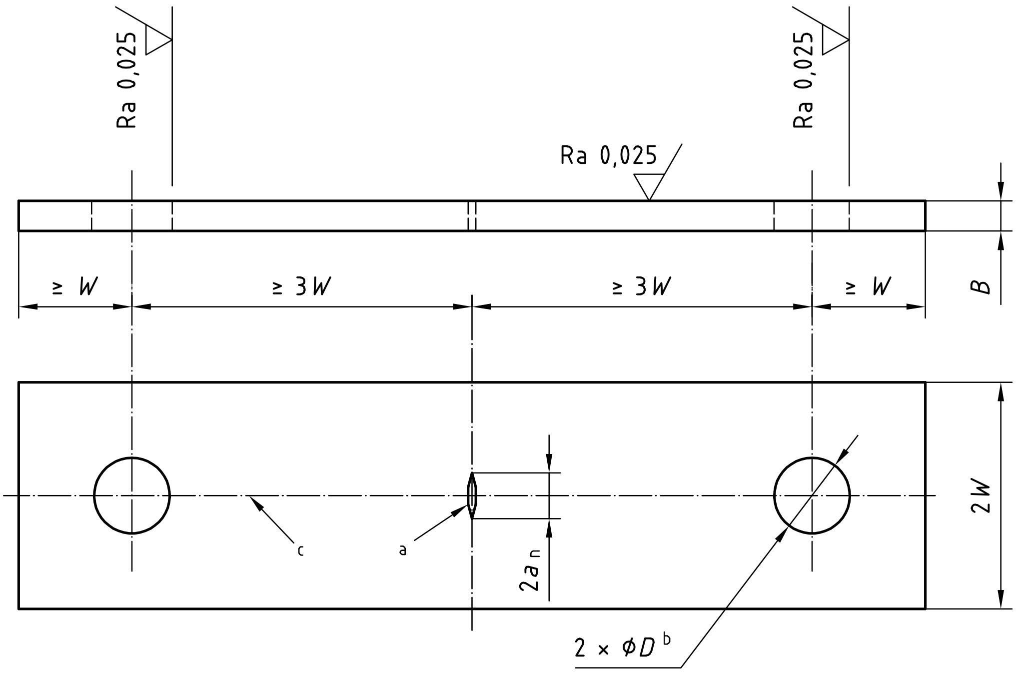
Figure 2 — Standard single edge notch tension, SENT, specimen

5.4.2 Tension-tension testing of a CCT specimen

For tension-tension testing of a specimen with a width $2W$, less than 75 mm, as shown in Figure 3, a clevis with single force pin is acceptable for gripping provided the specimen gauge length, defined here as the distance between the pin hole centrelines, be at least $6W$. Shims may be helpful in circumventing fretting fatigue at the specimen's pin hole. Another step that can be taken to prevent crack initiation at the pin holes is the welding or adhesive bonding of reinforcement plates or tabs to the gripping area, especially when testing very thin materials. Cutting the test section down in width to form a “dog bone” shaped specimen design is another measure that can be adopted to circumvent failure at the pin holes; here the gauge length is defined as the uniform width section and it shall be at least 3,4W in length.

For tension-tension testing of a specimen with a width greater than 75 mm, distributing the force across the specimen width with multiple pin holes is recommended. A serrated grip surface at the specimen-grip interface increases the force that can be transferred. With this force application arrangement the gauge length between the innermost rows of pin holes must be at least $3W$.

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,002W.

NOTE 2 The faces are parallel to $\pm$ 0,05 mm/mm.

NOTE 3 The two faces are not out of plane more than 0,05 mm.

NOTE 4 The crack length is measured from the reference plane of the longitudinal centreline.

NOTE 5 The clevis and pin loading system is not suitable for a force ratio $R < 0$.

NOTE 6 Special gripping systems may be used for a force ratio $R < 0$ such as shown in Figure 4.

[a] See Figure 12 for notch detail.

[b] $D = 2W/3$.

[c] Reference plane.

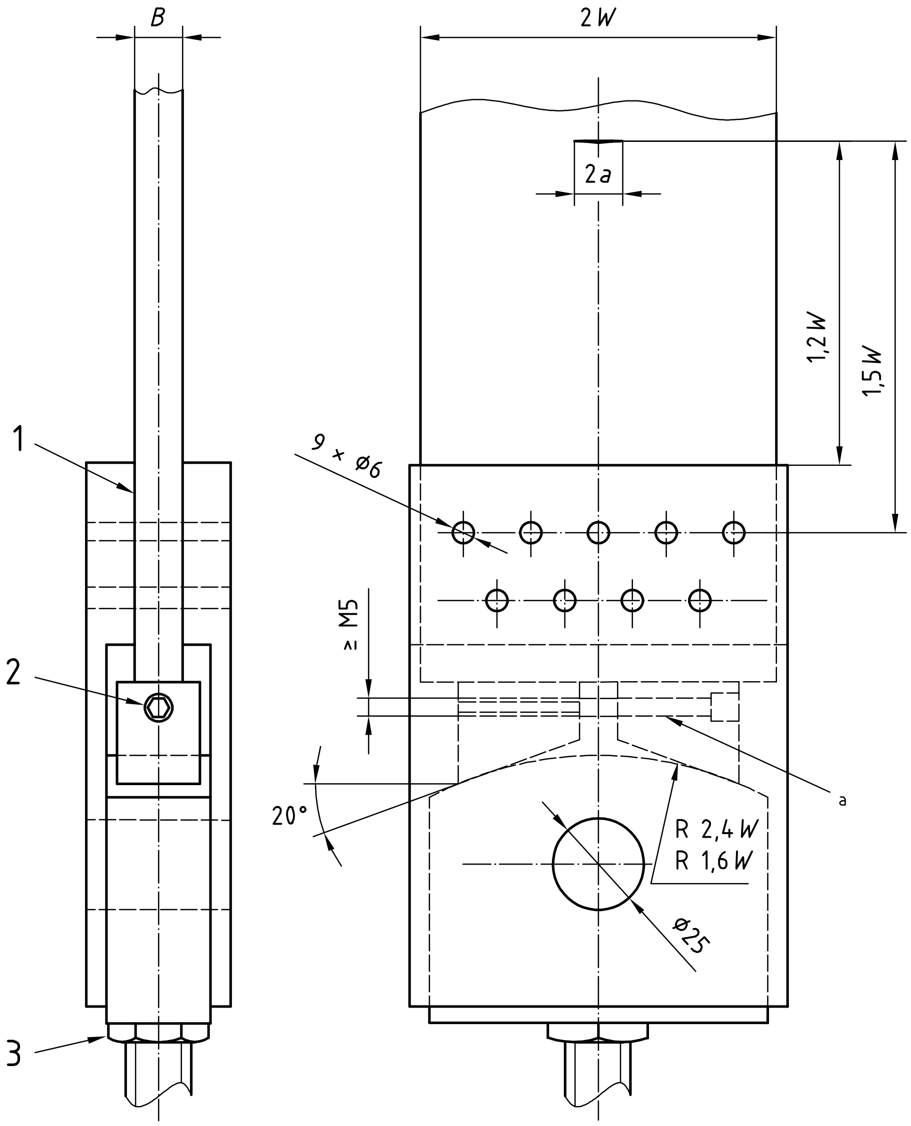
Figure 3 — Standard pinned end centre cracked tension, CCT, specimen for $2W \leqslant 75$ mm

5.4.3 Tension-compression testing of a CCT specimen

A backlash-free gripping arrangement shall be used for tension-compression testing of the CCT specimen. Various commercially available pneumatic and hydraulic wedge grips that provide adequate clamping force may be used. The minimum gauge length for a clamped CCT specimen is 2,4W.

For tension-compression testing of a CCT specimen, Figure 4 presents a design that affords a simple backlash free grip that provides improved force transfer through multiple pins plus frictional force transfer via specimen clamp-up with the serrated gripping surfaces. The compressive condition between the pins and the specimen's end surfaces, induced by drawing the wedges together, affords large reverse force excursions while circumventing elongation of the pin holes. The minimum gauge length for this specimen is 2,4W between the grip end surfaces and 3W between the inner rows of pins, as stated above.

Dimensions in millimetres



Key

1 Serrated sideplate surface

2 Countersunk cap screw

3 Lock nut

NOTE 1 Made of hardened steel, e.g. $\geqslant$ 40 HRC.

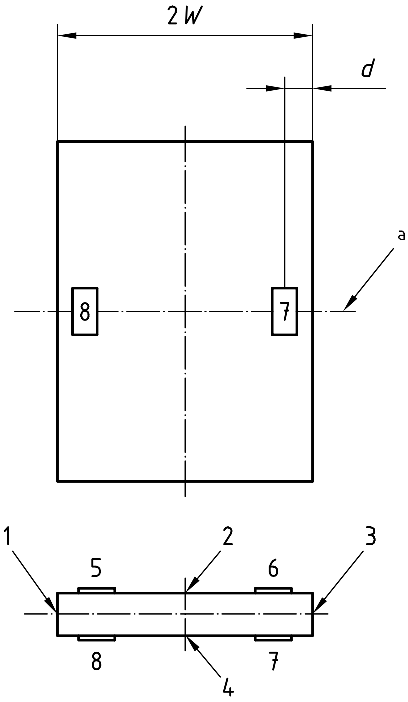
NOTE 2 Serrated side plates vary in thickness to accommodate approximately 2 mm to 3 mm, range in thickness B.

a Body drilled.

Figure 4 — Example of backlash free grip for a CCT specimen

5.4.4 Alignment of CCT specimen grips

The CCT specimen is sensitive to misalignment and nonsymmetrical force application, especially in tension-compression testing where gimbaled connections are not used, which can readily lead to violation of the through thickness crack curvature and/or symmetry validity criteria. It is recommended that bending strain be checked periodically with a panel specimen similar to the one being tested and instrumented with strain gauges, as shown in Figure 5 [22]. This technique can be used to minimize the bending strain. See 5.1.2.



The average axial strain, ϵ_a, for the flat panel calibration specimen is calculated using:

$$\epsilon_a = \frac{(\epsilon_5 + \epsilon_6 + \epsilon_7 + \epsilon_8)}{4}$$

where ϵ_5, ϵ_6, ϵ_7 and ϵ_8 are the measured strains.

The equivalent strain at the centre of the four faces 1 to 4 is calculated using:

$\epsilon_1 = \epsilon_a - [\epsilon_a - (\epsilon_5 + \epsilon_8)/2][2W/(2W - 2d)]$;

$\epsilon_3 = \epsilon_a - [\epsilon_a - (\epsilon_6 + \epsilon_7)/2][2W/(2W - 2d)]$;

$\epsilon_2 = (\epsilon_5 + \epsilon_6)/2$; $\epsilon_4 = (\epsilon_7 + \epsilon_8)/2$.

The local bending strains at the centre of each of the four faces are calculated using:

$b_1 = \epsilon_1 - \epsilon_a$; $b_2 = \epsilon_2 - \epsilon_a$;

$b_3 = \epsilon_3 - \epsilon_a$; $b_4 = \epsilon_4 - \epsilon_a$.

The maximum bending strain percentage in plane A can then be calculated as follows:

$$\beta\,\% = [(b_1 - b_3)/2 + (b_2 - b_4)/2]\,100/\epsilon_a \leqslant 5\,\%$$

[a] Plane A

Figure 5 — Strain gauge arrangement and bending strain calculation for an instrumented panel alignment specimen [22]

5.5 Grips and fixtures for the SENB specimens

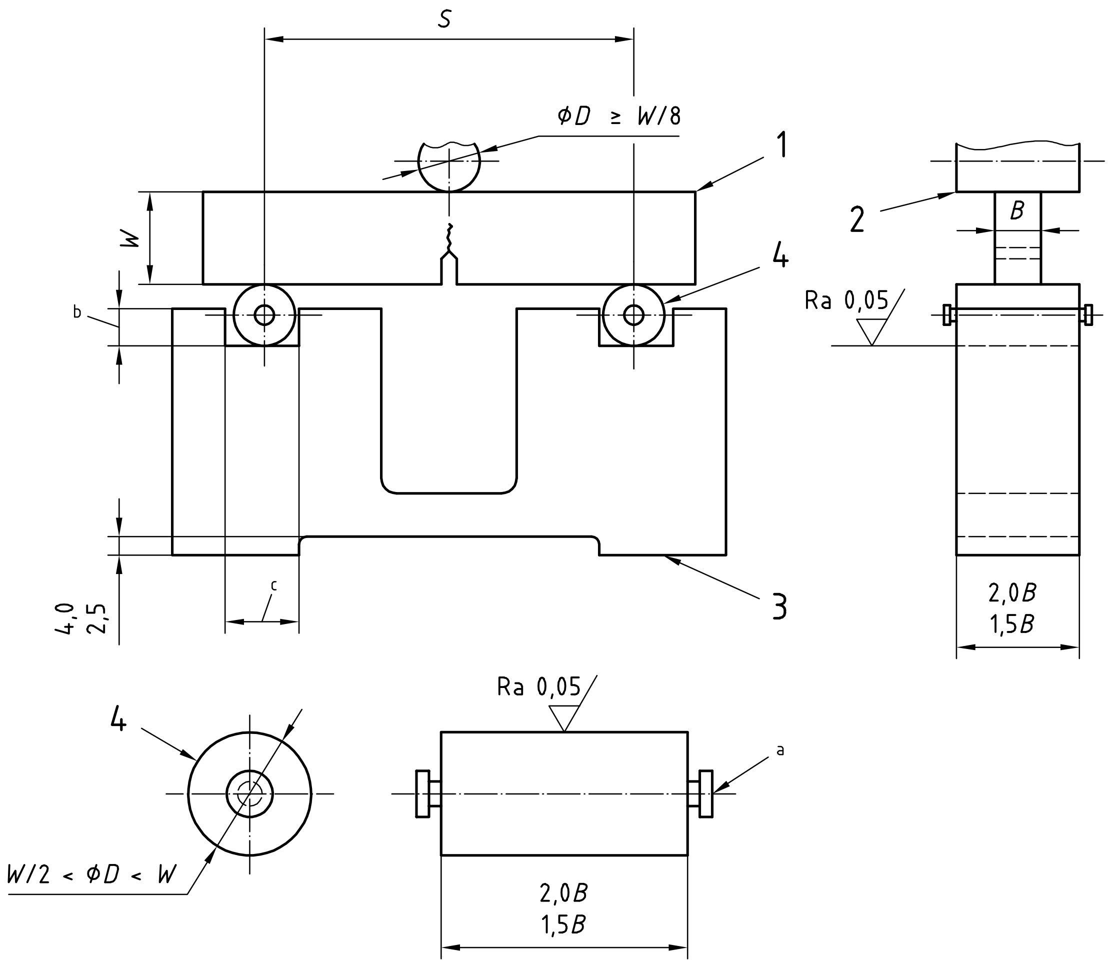
5.5.1 Tension-compression grips for the SENB8 specimen

The eight-point bend specimen is also suited for tension-compression testing. In gripping the eight-point bend specimen, the top and bottom tups are rigidly tied together with a line-to-line fit to the specimen's surfaces. Precautions shall be taken to eliminate backlash and secondary moments.

5.5.2 Tension-tension testing of SENB specimens

The general principles of the bend test fixture suitable for tension-tension testing of the SENB specimen are illustrated in Figure 6. The fixture is designed to minimize frictional effects by allowing the support rollers to rotate and move apart slightly as force is applied to the specimen, hence permitting rolling contact. Thus, the support rollers are allowed limited motion along plane surfaces parallel to the notched side of the specimen, but are initially positively positioned against stops that set the span length and are held in place by low-tension springs (such as rubber bands). Fixtures and rollers shall be made of high hardness ($>$ 40 HRC) steel [23].

Dimensions in millimetres
Surface roughness values in micrometres



Key

1 Test specimen

2 Ram

3 Test fixture

4 Roller pin

NOTE Roller pins and specimen contact surface of load ram should be parallel to each other to $\pm$ 0,002W (TIR[d])

[a] Bosses for springs or rubber bands.

[b] 0,6$\times$ roller pin diameter.

[c] 1,1$\times$ roller pin diameter.

[d] Total indicated reference value.

Figure 6 — Fixture for tension-tension forcing of a SENB3 specimen

5.6 Crack length measurement apparatus

5.6.1 General

Accurate measurement of crack length during the test is very important. There are a number of visual and non-visual apparati that can be used to determine the crack length. A brief description of a variety of crack length measurement methods is included in [26]. The required crack length measurements are the average of the through-the thickness crack lengths, as covered in 9.1.

5.6.2 Non-visual crack length measurement

There are a number of non-visual measurement techniques. Most lend themselves to automated data acquisition and determine the average crack length, reflecting the crack front curvature if it exists. Crack-opening-displacement compliance [36]-[38], AC and DC electric potential difference (EPD) [39]-[41], back face strain [36], [42], and side face foil crack gauges [43]-[45] are all acceptable techniques, provided the resolution requirements covered in 8.1 be met.

5.6.3 Visual crack length measurement

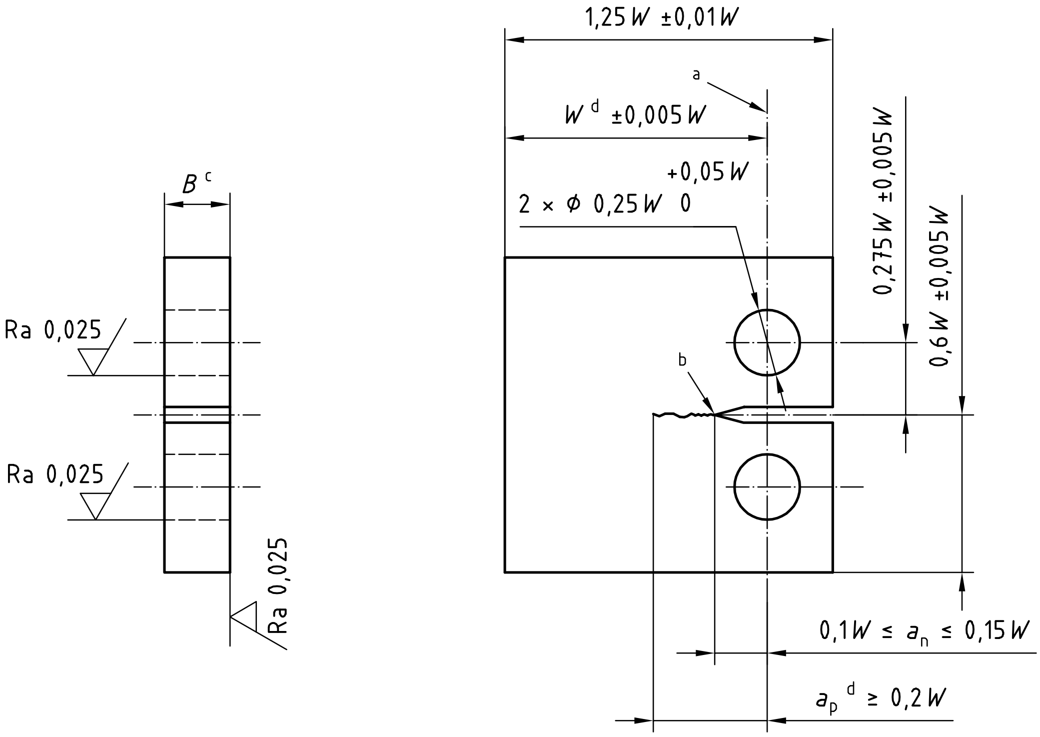
In the past, the most common visual crack length measurement technique used a micrometer thread travelling microscope with low magnification ($\times$20 to $\times$50). This technique measures the surface crack length during the test and may need to be corrected to the actual through-thickness crack size upon test completion, as covered in 9.1.

6 Specimens

6.1 General

Proportional dimensions of six standard specimens: a compact tension (CT); a centre cracked tension (CCT) and three, four and eight point single edge notch bend (SENB3), (SENB4) and (SENB8); and single edge notch tension (SENT) are presented in Figures 7, 3, 8, 9, 10 and 2, respectively. A variety of specimen configurations is presented to accommodate the component geometry available and test environment and/or force application conditions during a test. Machining tolerances and surface finishes are also given in Figures 7 to 10. The CT, SENB3 and SENB4 specimens are recommended for tension-tension test conditions only.

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,005W.

NOTE 2 The surfaces are perpendicular and parallel to within $\pm$ 0,002W (TIR).

NOTE 3 The crack length is measured from the reference plane of the loading pin holes centreline.

NOTE 4 Specimens recommended for notch root tension loading, with a force ratio $R > 0$, only.

[a] Reference plane.

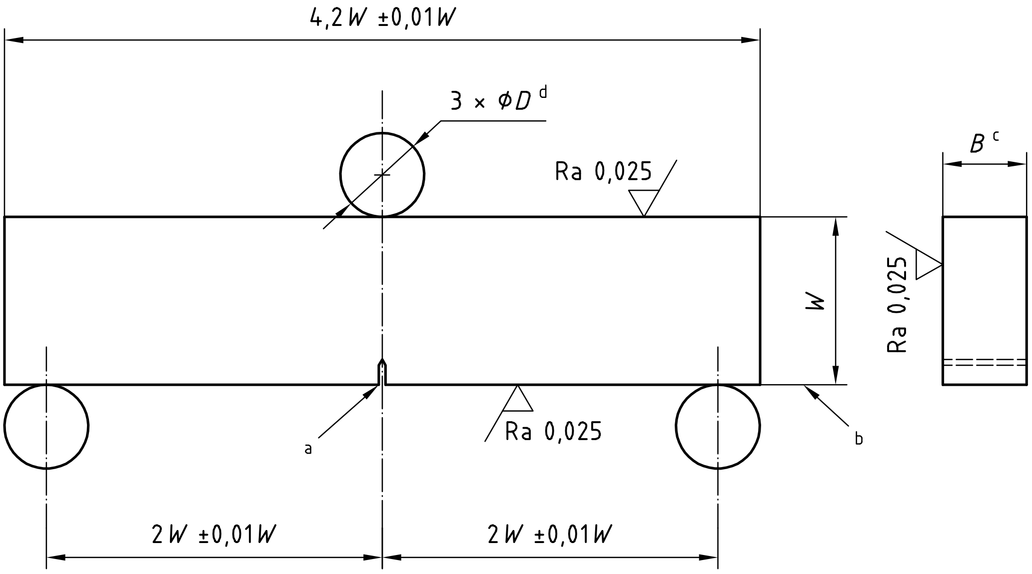
[b] See Figure 12 for notch detail.

[c] The recommended thickness: $W/20 \leqslant B \leqslant W/2$.

[d] The suggested minimum dimensions are $W = 25$ mm and $\alpha_p = 0{,}2W$.

Figure 7 — Standard compact tension, CT, specimen for fatigue crack growth rate testing

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,005W.

NOTE 2 The surfaces are parallel and perpendicular to within $\pm$ 0,002W.

NOTE 3 The crack length is measured from the reference loading plane containing the starter V-notch.

NOTE 4 Specimen recommended for notch root tension only.

[a] See Figure 12 for notch detail.

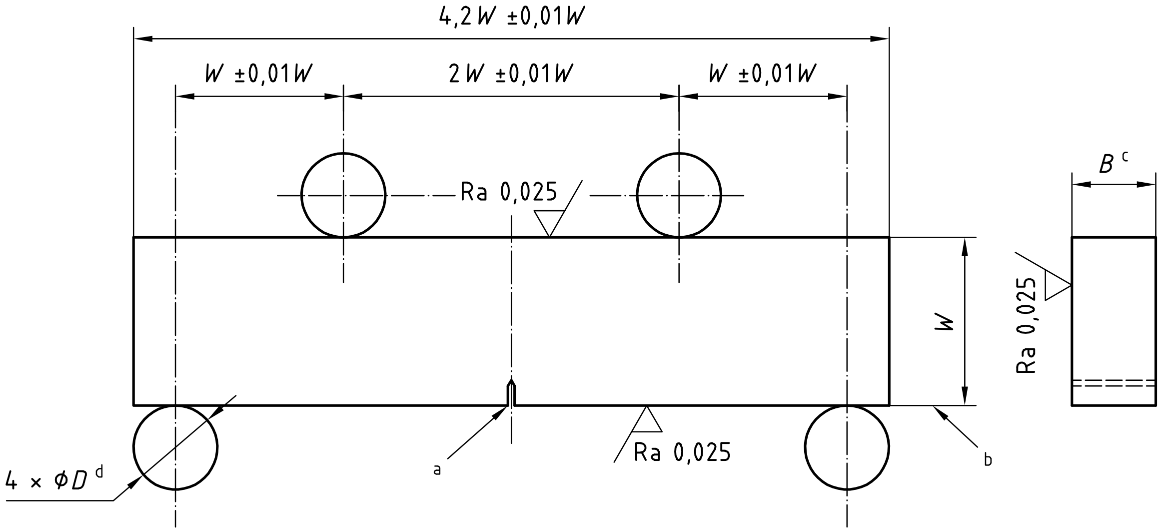
[b] Reference plane.

[c] Recommended thickness: 0,2$W \leqslant B \leqslant W$.

[d] $D \geqslant W/8$.

Figure 8 — Standard three point single edge notch bend, SENB3, specimen

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,005W.

NOTE 2 The surfaces are parallel and perpendicular to $\pm$ 0,002W (TIR).

NOTE 3 The crack length is measured from the reference loading plane containing the V-notch.

NOTE 4 Specimen recommended for tension $R \geqslant 0$ only.

[a] See Figure 12 for notch detail.

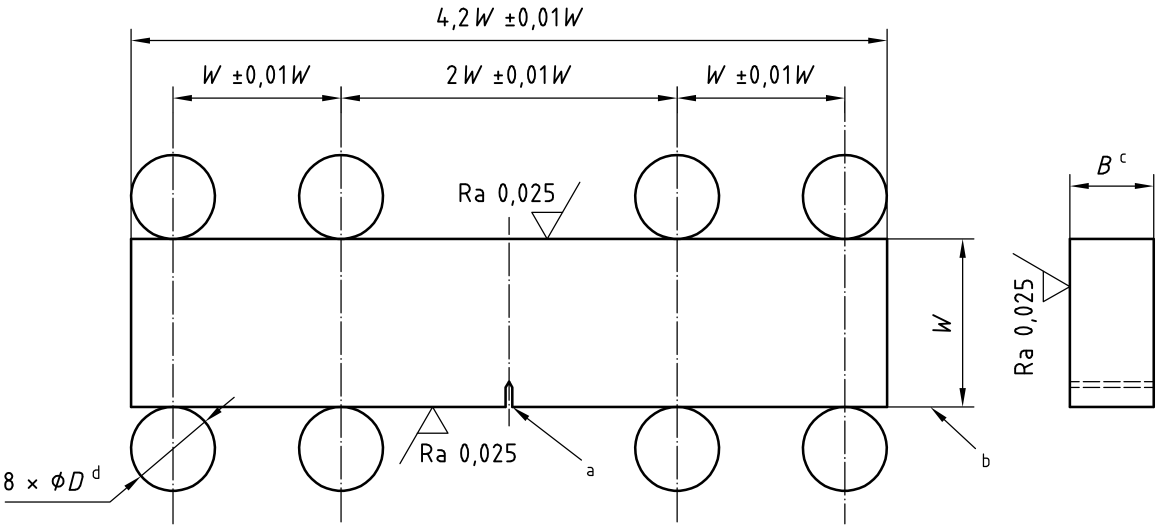
[b] Reference plane.

[c] Recommended thickness: 0,2$W \leqslant B \leqslant W$.

[d] $D \geqslant W/8$.

Figure 9 — Standard four point single edge notch bend, SENB4, specimen

Surface roughness values in micrometres



NOTE 1 The machined notch is centred to within $\pm$ 0,005W.

NOTE 2 The surfaces are parallel and perpendicular to within $\pm$ 0,002W (TIR).

NOTE 3 The crack length is measured from the reference loading plane containing the V-notch.

NOTE 4 Specimen suitable for $R \leqslant$ 0, provided backlash and secondary moment loading by grips be avoided.

[a] See Figure 12 for notch detail.

[b] Reference plane.

[c] The recommended thickness is 0,2$W \leqslant B \leqslant W$.

[d] $D \geqslant W/$8.

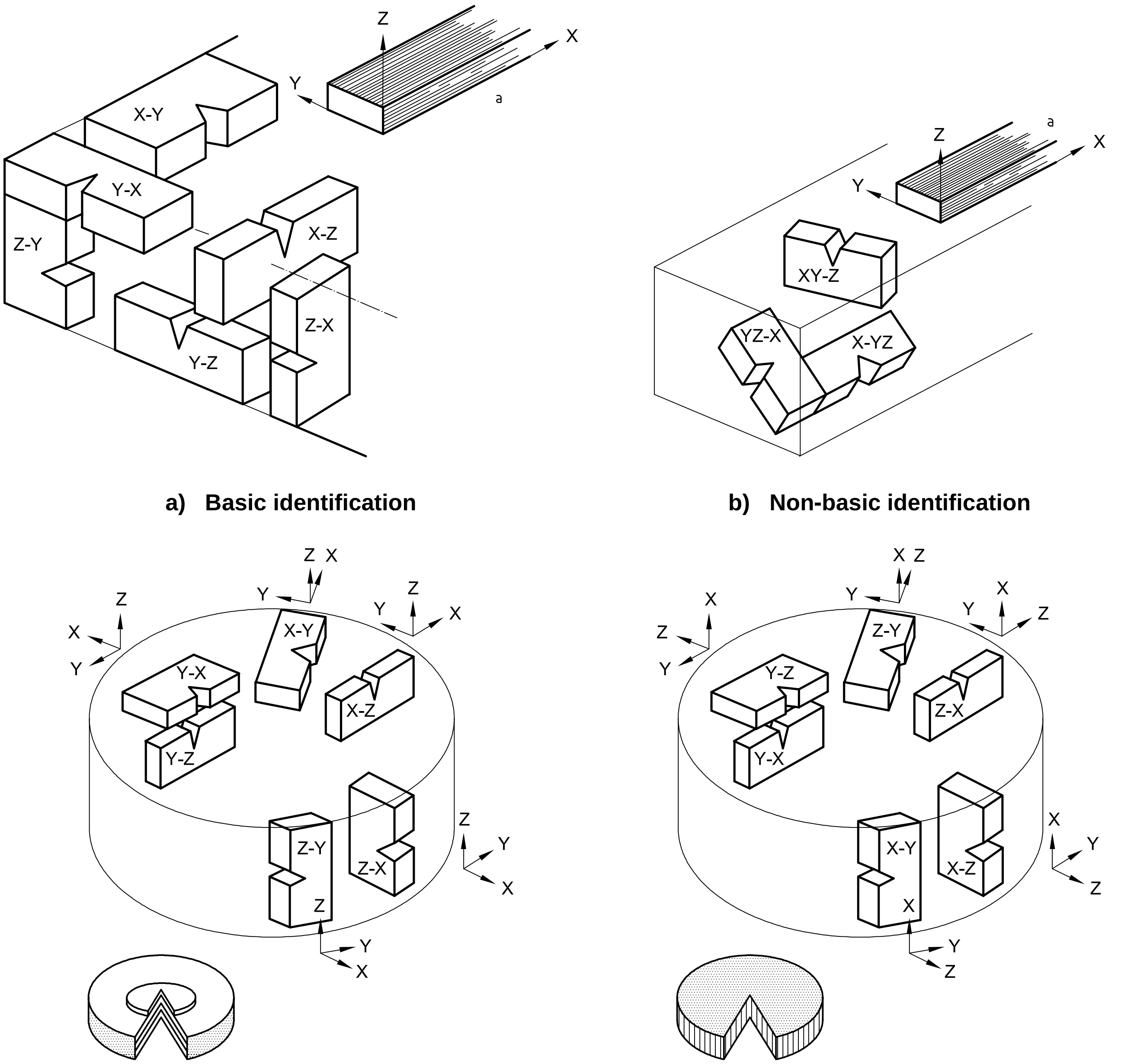
Figure 10 — Standard eight point single edge notch bend, SENB8, specimen

6.2 Crack plane orientation

The crack plane orientation, as related to the characteristic direction of the product, is identified in Figure 11. The letter(s) preceding the hyphen represent the force direction normal to the crack plane; the letter(s) following the hyphen represent the expected direction of crack extension. For wrought metals the letter X always denotes the direction of principal processing deformation, Y denotes the direction of least deformation and the letter Z is the third orthogonal direction. If the specimen orientation does not coincide with the product's characteristic direction then two letters are used before and/or after the hyphen to identify the normal to the crack plane and/or expected direction of crack extension.

The specimen shall have the same metallurgical structure as the material for which the crack growth rate is being determined. No heat treatment shall take place once the fatigue crack has been initiated.

The starter notch for the standard specimens may be made via electrical discharge machining (EDM), milling, broaching or saw cutting. To facilitate precracking, the notch root radius shall be as small as practical, typically less than 0,2 mm. For aluminium, saw cutting the final 0,5 mm starter notch depth with a jeweller's saw is acceptable.



a) Basic identification **b) Non-basic identification**

c) Radial grain flow, axial working direction **d) Axial grain flow, radial working direction**

[a] Grain flow.

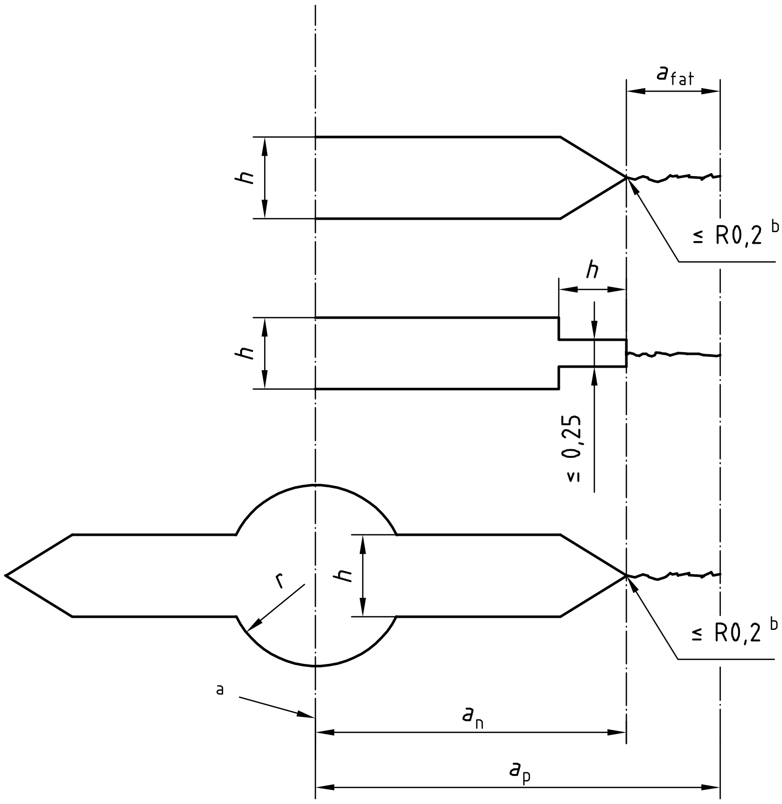
Figure 11 — Fracture plane orientation identification as in ISO 12737

6.3 Starter notch precracking details

The envelope and various acceptable machined notch configurations and precracking details for the specimens are presented in Figure 12. For all specimens, the notch height, h, shall not exceed 1 mm for $W \leqslant 25$ mm, or $W/16$ for $W > 25$ mm. The machined notch shall be as narrow as practical machining limitations permit in order to minimize the precracking length requirement. It is required that the final precrack length, a_p, in the CT specimen be at least 0,2W in length in order to avoid invalidating the K-calibration expression.

The machined notches in the SENB and CCT specimens are determined by practical machining limitations; the K-calibration does not have a notch size limitation. However, a CCT specimen's minimum precrack length, 2a_p, of at least 0,4W is required when using the compliance method for crack length determination to assure accurate crack length measurements.

Dimensions in millimetres



NOTE 1 Crack length is measured from reference plane.

NOTE 2 Notch height, h, should be minimized.

NOTE 3 A hole of radius $r < 0{,}05W$ is allowed for ease of machining the notch in a CCT specimen.

[a] Reference plane.

[b] Root radius.

Specimen type	Notch length a_n	Maximum notch height h	Minimum precrack length a_p
CT CCT	$0{,}1W \leqslant a_n \leqslant 0{,}15W$	$\leqslant 1$ for $W \leqslant 25$ $W/16$ for $W > 25$	$a_p \geqslant a_n + h$ $a_p \geqslant a_n + 1$ mm $a_p \geqslant a_n + 0{,}1B$, and $a_p \geqslant 0{,}2W$ for CT only

Figure 12 — Notch detail and minimum fatigue precracking requirements

6.4 Stress intensity factor

6.4.1 General

The stress intensity factor for all standard specimen configurations is calculated using the following relationship:

$$K = \frac{F}{BW^{1/2}}\, g\left(\frac{a}{W}\right) 10^{1{,}5} \tag{1}$$

The stress intensity factor function, $g\left(a/W\right)$, for each standard specimen configuration is calculated using the following expressions.

6.4.2 Compact tension, CT, specimen

For a compact tension specimen, CT, the stress intensity factor function is given by

$$g\left(\frac{a}{W}\right) = \frac{(2+\alpha)\left(0{,}886 + 4{,}64\alpha - 13{,}32\alpha^2 + 14{,}72\alpha^3 - 5{,}6\alpha^4\right)}{(1-\alpha)^{3/2}} \tag{2}$$

where $\alpha = a/W$; the expression is valid for 0,2 $\leqslant a/W \leqslant$ 1,0. See Figure 7.

6.4.3 Centre cracked tension, CCT, specimen

For the centre cracked tension specimen, CCT, the stress intensity factor function is given by [24]-[26]:

$$g\left(\frac{a}{W}\right) = \left(\frac{\theta}{\cos\theta}\right)^{1/2} \left(0{,}707\,1 - 0{,}007\,2\theta^2 + 0{,}007\,0\theta^4\right) \tag{3}$$

where $\theta = \pi a/2W$ radians; the expression is valid for $0 < \alpha = 2a/2W < 1{,}00$. Here it is recommended that the crack length, a, be the average of the four measurements from the centreline reference plane to the crack tips on both the front and back surfaces. See Figure 3.

6.4.4 Single edge notch three point bend, SENB3, specimen

For the three point bend specimen, SENB3, with the distance between external supports equalling $4W$, the stress intensity factor function is given by [27]:

$$g\left(\frac{a}{W}\right) = \frac{6\alpha^{1/2}}{\left[(1+2\alpha)(1-\alpha)^{3/2}\right]} \left[1{,}99 - \alpha(1-\alpha)\left(2{,}15 - 3{,}93\alpha + 2{,}7\alpha^2\right)\right] \tag{4}$$

where $\alpha = a/W$; the expression is valid for $0 \leqslant \alpha \leqslant 1{,}0$. See Figure 8.

6.4.5 Single edge notch four point bend, SENB4, specimen

For the four point bend specimen, SENB4, with the distance between external supports minus the distance between internal supports equalling $2W$, the stress intensity geometric function is given by [10]:

$$g\left(\frac{a}{W}\right) = 3\left(2\tan\theta\right)^{1/2} \left[\frac{0{,}923 + 0{,}199\left(1-\sin\theta\right)^4}{\cos\theta}\right] \tag{5}$$

where $\theta = \pi a/2W$ radians; the expression is valid for $0 \leqslant a/W \leqslant 1{,}0$. See Figure 9.

For the four point bend specimen, where the difference between the major and minor span does not equal $2W$, the value for $g(1/W)$ is proportional to the ratio of the major span minus the minor span divided by $2W$,

i.e. $\dfrac{(\text{major span} - \text{minor span})}{2W}$

6.4.6 Single edge notch eight point bend, SENB8, specimen

For the eight point bend specimen, SENB8, with the distance between external supports minus the distance between internal supports equaling $2W$, the stress intensity factor function is given by [10]:

$$g\left(\frac{a}{W}\right) = 3\left(2\tan\theta\right)^{1/2}\left[\frac{0{,}923 + 0{,}199\left(1-\sin\theta\right)^4}{\cos\theta}\right]$$

where $\theta = \pi a/2W$ radians; the expression is valid for $0 \leqslant a/W \leqslant 1{,}0$. See Figure 10.

For the eight point bend specimen, where the difference between the major and minor span does not equal $2W$, the value for $g\left(a/W\right)$ is proportional to the ratio of the major span minus the minor span divided by $2W$.

i.e. $\dfrac{\text{major span} - \text{minor span}}{2W}$

6.4.7 Single edge notch tension, SENT, specimen

For the single edge notch pinned end tension specimen, SENT, the stress intensity factor function is given by [10]:

$$g\left(\frac{a}{W}\right) = \sqrt{2\tan\theta}\left[\frac{0{,}752 + 2{,}02\alpha + 0{,}37\left(1-\sin\theta\right)^3}{\cos\theta}\right] \qquad (6)$$

where $\theta = \pi a/2W$; the expression is valid for $0 < a/W < 1{,}0$. See Figure 2.

For the single edge notch clamped end tension specimen, with the clear span between the grips equalling $4W$, the stress intensity factor function is given by [28]:

$$g\left(\frac{a}{W}\right) = \left(1-\alpha\right)^{-3/2}\left[1{,}987\,8\alpha^{1/2} - 2{,}972\,6\alpha^{3/2} + 6{,}950\,3\alpha^{5/2} - 14{,}447\,6\alpha^{7/2} + 10{,}054\,8\alpha^{9/2} + 3{,}404\,7\alpha^{11/2} - 8{,}714\,3\alpha^{13/2} + 3{,}741\,7\alpha^{15/2}\right] \qquad (7)$$

where $\alpha = a/W$; the expression is valid for $0 < a/W \leqslant 0{,}95$. See Figure 2.

Stress intensity factor functions, for clamped end SENT specimens with spans between the grips other than $4W$, are available [29]-[31].

6.5 Specimen size

6.5.1 General

For the test results to be valid it is required that the specimen remain predominantly in a linear-elastic stress condition throughout the test. The specimen width, W, and thickness, B, may be varied independently within the limits covered in 6.6. The smallest specimen to meet these criteria, based on experimental results, varies with each specimen configuration [32].

The minimum uncracked ligament that circumvents large scale yielding varies with specimen configuration and is a function of the material's 0,2 % proof strength.

6.5.2 CT specimen minimum uncracked ligament

For the CT specimens the minimum uncracked ligament for producing valid data is given by:

$$(W-a) \geqslant \left(\frac{4\,000}{\pi}\right)\left(\frac{K_{\text{max}}}{R_{\text{p0,2}}}\right)^2 \quad (8)$$

6.5.3 CCT specimen minimum uncracked ligament

For the CCT specimen the minimum size of the uncracked ligament, based upon large scale net section yielding of the material, is given by:

$$2\,(W-a) \geqslant \frac{1\,250 F_{\text{max}}}{B R_{\text{p0,2}}} \quad (9)$$

6.5.4 SENB specimen minimum uncracked ligament

For all of the bend SENB specimens the minimum size of the uncracked ligament is given by:

$$(W-a) \geqslant \left(\frac{3\,000 \lambda F_{\text{max}}}{2 B R_{\text{p0,2}}}\right)^{0,5} \quad (10)$$

This criterion is based upon large scale net section yielding of the material and $\lambda = 4W$, the distance between external supports for a three point bend specimen; $\lambda = 2W$ for a four and eight point bend specimen or if a nonstandard four or eight point bend specimen is used, λ equals the distance between external supports minus the distance between internal supports.

6.5.5 SENT specimen minimum uncracked ligament

The minimum size of the uncracked ligament for the SENT specimen depends on the gripping technique: for a tensile stress, e.g. the ends embedded in hydraulic wedge grips, the minimum uncracked ligament is given by:

$$(W-a) \geqslant \frac{1\,250 F_{\text{max}}}{B R_{\text{p0,2}}} \quad (11)$$

This criterion is based upon large scale net section yielding of the material.

For a bending stress, e.g. clevis and pinned end grips, the minimum uncracked ligament is given by equation (8).

6.6 Specimen thickness

6.6.1 General

Specimen thickness, B, may be varied independent of specimen width, W, for the specimen configurations, within the limits for buckling and through-thickness crack front curvature considerations. It is recommended that the selected specimen thickness be similar to that of the product under study.

6.6.2 CT specimen

For a CT specimen it is recommended that the thickness, B, be within the range $W/20 \leqslant B \leqslant W/4$. A thickness up to $W/2$ is permitted. For a specimen this thick, a through-thickness crack front curvature correction length, a_{cor}, may often be required; also, difficulties may be encountered in meeting the through-thickness crack straightness requirements covered in 9.1.

6.6.3 CCT specimen

For the CCT specimen the recommended upper limit for thickness is $W/8$, although $W/4$ is permitted. The minimum thickness for circumventing out-of-plane deflection or buckling in the CCT specimen is dependent on the test material's elastic modulus, E, gauge length, gripping, grip alignment and force ratio, R.

6.6.4 SENB specimen

For the single edge notch bend specimen it is recommended that the thickness be within the range $0{,}2W \leqslant B \leqslant W$.

6.6.5 SENT specimen

For the single edge notch tension specimen the maximum recommended thickness equals $0{,}5W$.

6.7 Residual stresses

Residual stresses in a material that has not been stress relieved can influence the crack propagation rate considerably [12], [13]. This influence can be minimized by choosing a symmetrical specimen configuration like the standard CCT specimen and reducing the B/W ratio to minimize crack front curvature caused by variation in residual stresses through the thickness [13].

7 Procedure

7.1 Fatigue precracking

The purpose of precracking is to provide a straight and sharp fatigue crack of sufficient length so that the K-calibration expression is no longer influenced by the machined starter notch and that the subsequent fatigue crack growth rate is not influenced by a changing crack front shape or precracking force history.

The precrack length, a_{p}, shall equal or exceed $0{,}2W$ for the CT specimens only. The CCT specimen has no length requirement except when using compliance crack length measurement which necessitates $2a_{\text{p}} \geqslant 0{,}4W$. For all specimens the minimum fatigue precracking extension, a_{fat}, from the notch on each surface shall exceed the greater of the notch height, h, 1 mm or $0{,}1B$, as shown in Figure 12.

Prior to precracking, the test specimen shall be in the fully machined condition and in the final heat-treated state that the material will see in service. One practice is to initiate the fatigue crack at the lowest possible maximum stress intensity factor, K_{max}, that is practical. If the test material's critical stress intensity factor, which will cause fracture, is approximately known then the initial K_{max} for precracking can range from 30 % to 60 % of that value. If crack initiation does not occur within a block 30 000 to 50 000 load cycles then K_{max} can be increased by 10 % and the block of load cycles repeated. The final K_{max} for precracking shall not exceed the initial K_{max} for which test data are to be generated.

Frequently, a stress intensity factor, greater than the K_{max} used in the test, needs to be used for crack initiation. In this case, the maximum force shall be stepped down to meet the above criteria. When manually controlling precracking, the recommended stress intensity factor drop for each step is less than 10 % of K_{max}. In addition, it is recommended that between each stress intensity factor reduction that the crack extend by at least the value given in equation (12) [18]:

$$\Delta a_j = \frac{3}{\pi}\left[\frac{K_{\text{max}}\,(j-1)}{R_{\text{p0,2}}}\right]^2 \qquad (12)$$

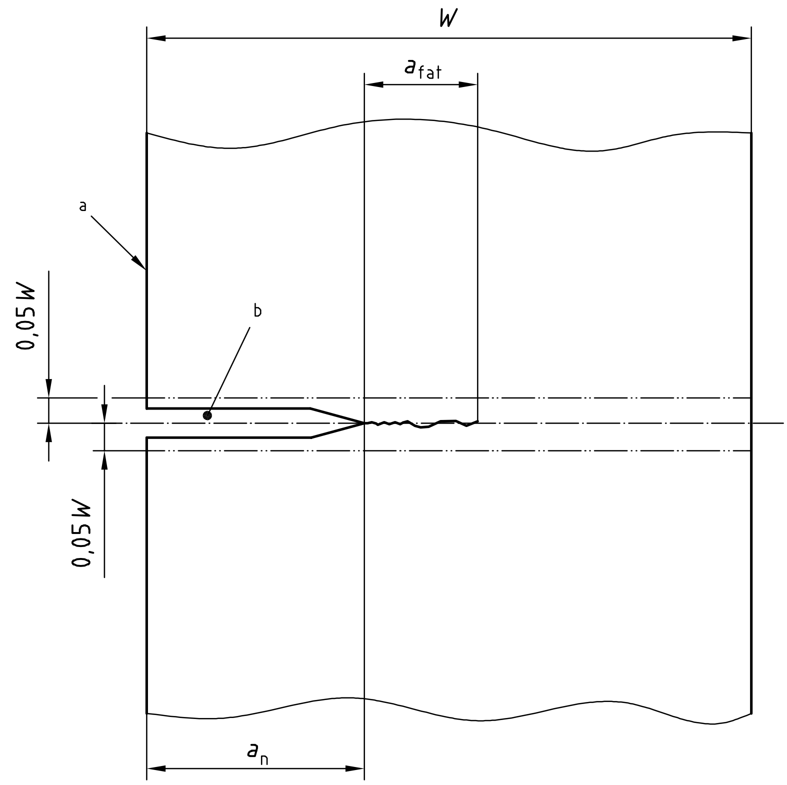
where $K_{\text{max}}\,(j-1)$ is the maximum terminal stress intensity factor of the previous step.

When test data are to be generated for a high force ratio it may be more convenient to precrack at a lower K_{max} and force ratio than the initial test conditions.

The precracking apparatus shall apply the force symmetrical to the specimen's notch and accurately maintain the maximum force to within 5 %. A centre cracked panel shall also be symmetrically stressed across the width, $2W$. Any frequency that accommodates maintaining the force accuracy specified in 5.1 is acceptable.

7.2 Crack length measurement

The requirements for measurement accuracy, frequency and validity are covered in clauses 8 and 9 for the various specimen configurations and test procedures that follow. When surface measurements are used to determine the crack length, it is recommended that both the front and back surface traces be measured. If the front to back crack length measurements vary by more than 0,25B and, for a CCT specimen, if the side-to-side symmetry of the two crack lengths vary in length by more than 0,025W then the precrack is not suitable and test data would be invalid under this test method. In addition, if the precrack departs from the plane of symmetry beyond the corridor, defined by planes 0,05W on either side of the specimen's plane of symmetry containing the notch root(s), the data would be invalid. See Figure 13.



[a] Reference plane.

[b] Machined notch, a_n.

Figure 13 — Out of plane cracking validity corridor

7.3 Constant-force-amplitude, K-increasing, test procedure for $da/dN > 10^{-5}$ mm/cycle

This procedure is appropriate for generating fatigue crack growth rate data above 10^{-5} mm/cycle. After stepping the maximum precracking force down to be equal or less than that corresponding to the lowest K_{max} in the range over which fatigue crack growth rate data will be generated, it is preferred that the force range be held constant as is the stress ratio and frequency. The maximum stress intensity factor will increase with crack extension and should be allowed to increase to equal or exceed the greatest K_{max} in the range over which data will be generated. Several suggestions, aimed at minimizing transient effects while using this K-increasing procedure, follow. If test variables are to be changed, $K_{\max}$ shall be increased rather than decreased in order to preclude the retardation effects attributable to the previous force history. Transient effects can also occur following a change in $K_{\min}$ or the stress ratio. An increase of 10 % or less in $K_{\max}$ and/or $K_{\min}$ will usually minimize the transient effect reflected in the fatigue crack growth rate. Following a change in force conditions sufficient crack extension shall be allowed to occur in order to re-establish a steady-state crack growth rate before the ensuing test data are accepted as valid under this test practice. The amount of crack extension required is dependent on many variables, e.g. percentage of force change, the test material and heat treatment condition. When environmental effects are present the amount of crack extension required to re-establish the steady-state growth rate may increase beyond that required in a benign environment.

Test interruptions shall be kept to a minimum. If the test is interrupted, a change in growth rate may occur upon resumption of cycling. The test data immediately following the interruption shall be considered invalid if there is a significant demarcation in the crack velocity from the steady-state growth rate immediately preceding the suspension of cycling. The sphere of influence of the transient effect may increase with the steady state force applied to the specimen during the suspension of dynamic force cycling.

7.4 K-decreasing procedure for $\mathrm{d}a/\mathrm{d}N < 10^{-5}$ mm/cycle

This K-decreasing procedure may result in different crack growth rates dependent on the test K-gradient, C. It is the user's responsibility to verify that the crack growth rates are not sensitive to the test K-gradient, C.

Testing starts at a $K_{\max}$ or stress intensity factor range, ΔK, equal to or greater than that used for the final crack extension while precracking. Following crack extension, the stress intensity factor range is stepped down, or continuously shed, at a constant rate until test data have been recorded for the lowest stress intensity factor range or fatigue crack growth rate of interest.

The K-decreasing test may be controlled by a stepped stress intensity factor following a selected crack extension at a constant ΔF, as shown in Figure 14. Alternately, the stress intensity factor gradient per increment of crack extension may be held constant, $1/\mathrm{d}a\,(\mathrm{d}K/K) =$ constant, called continuous stress intensity factor shedding, by using a computer automated test control procedure [46]; the constant, C, is called the normalized K-gradient. A common value is $C = -0{,}1\ \mathrm{mm}^{-1}$, but research has shown that this value may be material- and specimen-geometry dependent [47], [48]:

$$C = \left(\frac{1}{\mathrm{d}a}\right)\left(\frac{\mathrm{d}K}{K}\right) = \left(\frac{1}{K}\right)\frac{\mathrm{d}K}{\mathrm{d}a} \geqslant -0{,}1\ \mathrm{mm}^{-1} \tag{13}$$

This value usually provides a gradual enough force shed to preclude a transient in the crack growth rate. The above relationship is equally applicable whether calculating $K_{\max}$, $K_{\min}$ or ΔK, and can be rewritten for convenience in the integrated form as:

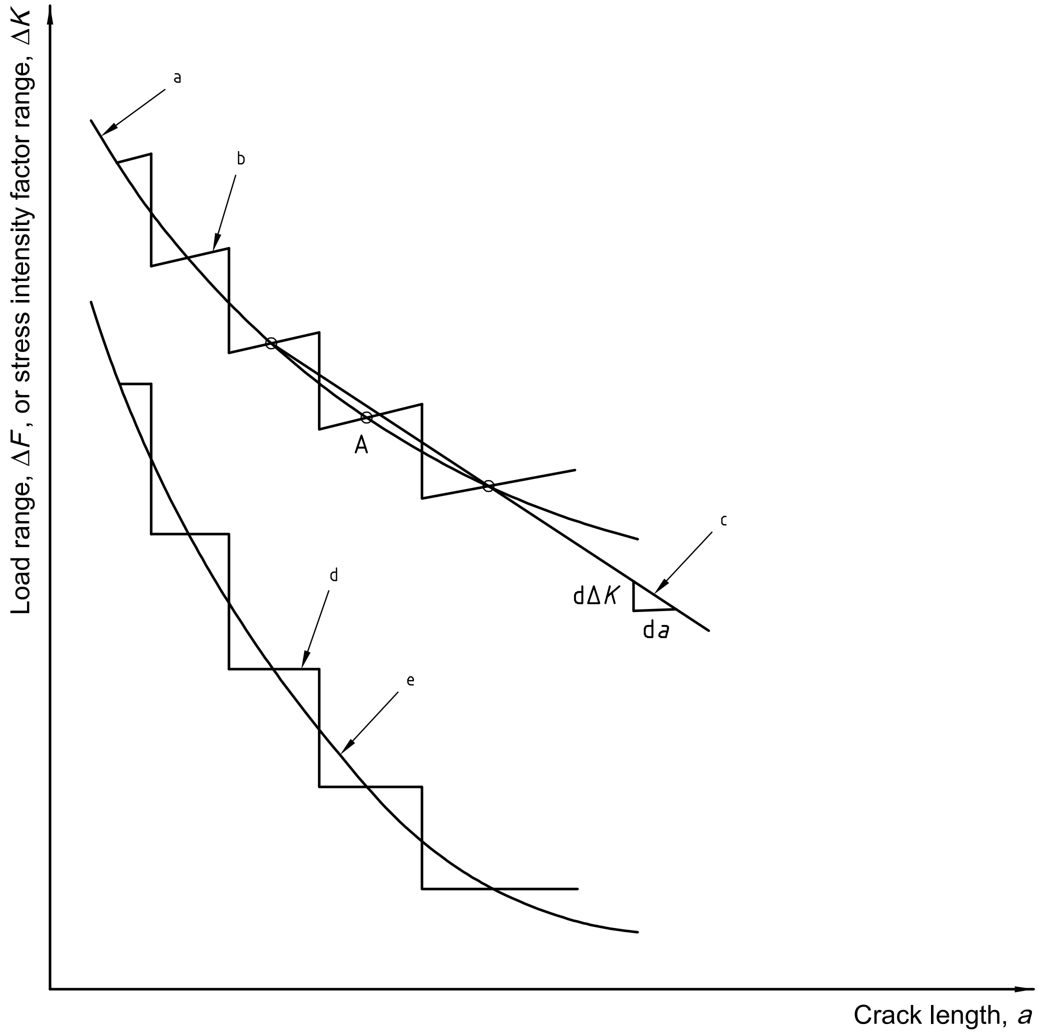
$$\Delta K_0\,(j) = \Delta K_0\,(j-1)\,\mathrm{e}^{C\Delta a(j-1)} \tag{14}$$

where

$\Delta K_0\,(j)$ and $\Delta K_0\,(j-1)$	are the initial stress intensity factor range at step j and $j-1$, respectively;
$\Delta a\,(j-1) = [a\,(j) - a\,(j-1)]$	is the crack extension at the preceding constant force range $\Delta F\,(j-1)$.

The stress ratio, R, and the normalized K-gradient, C, should be kept constant throughout the K-decreasing test. It is recommended that K-decreasing be followed by K-increasing testing procedure, as covered in 7.3.

When using the stress intensity factor stepped drop procedure, the reduction of $K_{\max}$ shall not exceed 10 % of the previous maximum stress intensity factor and a minimum crack extension $\Delta a \geqslant 0{,}50$ mm at each stress intensity step is recommended.



a ΔK nominal.

b ΔK actual.

c Slope $\approx$ nominal $\frac{\mathrm{d}\Delta K}{\mathrm{d}a}$ at point A.

d ΔF actual.

e ΔF nominal.

Figure 14 — Typical K-decreasing test by stepped force reduction method

When using continuous stress intensity shedding procedure the above requirement is inoperative. it is better to keep the force range constant for a very small crack extension, $\Delta a\,(j-1)$. Here, continuous stress intensity factor shedding is defined by the drop in initial stress intensity factor range, $\Delta K_\mathrm{i}\,(j)$, with each step, j, which may not exceed 2 % of the preceding initial stress intensity factor range, corresponding to:

$$\left[\frac{\Delta K_0\,(j-1)-\Delta K_0\,(j)}{\Delta K_0\,(j)}\right] \leqslant 0{,}02 \qquad (15)$$

E.g., if the common value $C = -$ 0,1 mm^{-1} is used, along with the maximum 2 % drop in each initial stress intensity range, then the exponent $C = \Delta a\,(j-1)$ and the crack extension for each constant force range equals $\Delta a\,(j-1) = 0{,}2$ mm:

$$\frac{\Delta K_0\,(j)}{\Delta K_0\,(j-1)} = \mathrm{e}^{C\Delta a(j-1)} \tag{16}$$

$$0{,}980 = \mathrm{e}^{(-0{,}1)(0{,}2)} = \frac{1}{2{,}718\,3^{0{,}02}} = \frac{1}{1{,}020\,2}$$

8 Crack length measurement

8.1 Resolution

The fatigue crack length measurements made as a function of elapsed force cycles may be made by techniques outlined in 5.6. The required resolution for crack length measurements is $\pm$ 0,1 mm or 0,002W, whichever is greater.

When making visual crack length measurements it is recommended that the surface in the area of the crack plane be polished and indirect lighting be used to enhance the visibility of the crack tip. It is highly recommended that crack length measurements be made on both the front and back faces of the specimen, to assure that crack symmetry requirements specified in 8.5 are met. The average of the surface crack length measurements, two for a CT, SENT and SENB specimens, and four for the CCT specimen, shall be used in calculating the crack growth rate and stress intensity factor range. If crack length is not measured on both faces for every crack length measurement then the interval between both front and back face measurements shall be reported. It is good practice to make regular comparisons between visual and non-visual measurement methods.

8.2 Interruption

Suspension of force cycling while making crack length measurements, although permitted, is discouraged and shall be avoided when possible. The duration and frequency of any interruptions should be kept to a minimum. Test interruption for making visual crack length measurements can be avoided by using strobe light illumination. See 7.3.

8.3 Static force

A static force may be maintained to enhance the resolution of the crack length measurements. A static force equal to or less than the fatigue mean force is usually acceptable. In corrosive or elevated temperature environments the mean force may introduce transient creep or blunting effects. In no case shall the applied static force exceed the maximum fatigue force. See 7.3.

8.4 Measurement interval

Crack length measurement shall be made so that da/dN data are uniformly distributed over the range of ΔK of interest. The following measurement intervals are recommended to provide a uniform data distribution:

— for the CT and SENB specimens,

$\Delta a \leqslant 0{,}04W$ for $0{,}25 \leqslant a/W < 0{,}40$

$\Delta a \leqslant 0{,}02W$ for $0{,}40 \leqslant a/W < 0{,}60$

$\Delta a \leqslant 0{,}01W$ for $a/W \geqslant 0{,}60$

— and for the CCT specimen,

$\Delta a \leqslant 0{,}03W$ for $2a/2W \leqslant 0{,}60$

$\Delta a \leqslant 0{,}02W$ for $2a/2W > 0{,}60$.

However, a minimum Δa of 0,25 mm is recommended. The above limits may need to be reduced in order to obtain multiple crack length measurements in the near threshold region. The minimum crack measurement interval in all cases must exceed ten times the crack length measurement precision. Here, precision is defined as the standard deviation from the mean value crack length determined for a set of repeat measurements.

8.5 Symmetry

As in 7.1, for any crack length measurement, the data are invalid if:

a) for a given crack front, the front and back crack length measurements differ by more than 0,25B, and

b) for a CCT specimen the symmetry of the two crack fronts differ by more than 0,05W then the crack is not suitable and the data are invalid by this test method.

When using a nonvisual method for crack length measurement the crack length should be visually checked for symmetry at the test start and finish, and at least three additional, evenly spaced, intermediate measurements are recommended.

8.6 Out-of-plane cracking

If the crack deviates from the theoretical crack plane by more than the 0,05W corridor, as covered in 7.2, the ensuing data are invalid. See Figure 13. Large grained or single-crystal materials can commonly violate this requirement for out-of-plane cracking.

8.7 Crack tip bifurcation

Crack front splitting or branching can be a source of variability in the measured fatigue crack growth rate data since it is not compensated for in the stress intensity factor calculation. When crack tip branching or bifurcation is present it shall be noted in the final report.

9 Calculations

9.1 Crack front curvature

After completion of the test the fracture faces shall be examined for through-thickness crack front curvature. Measure the crack lengths at the two specimen faces and the three-quarter-thickness crack lengths, i.e. at 0,25B, 0,50B and 0,75B from one of the faces; the average of the three-quarter-thickness measurements is called the average through-thickness crack length. The difference between the average through-thickness crack length and the corresponding crack length at the specimen faces during the test is the crack curvature correction length, a_{cor}. It is desirable to make the crack curvature correction calculation at more than one location on the fracture face where the fatigue crack front is clearly marked. If the crack curvature correction results in a more than 5 % difference in the calculated stress-intensity factor at any location then this correction must be included when analysing the recorded test data, and the effective crack length becomes:

$$a = a_{\text{n}} + a_{\text{fat}} + a_{\text{cor}} \tag{17}$$

When the magnitude of the crack curvature correction varies with crack length, a linear interpolation is used to determine the correction for the intermediate data.

9.2 Determining the fatigue crack growth rate

9.2.1 General

The fatigue crack growth rate is determined from the test record data pairs of crack length and corresponding elapsed force cycles. Two common methods used for calculating the crack growth rate, the secant method and incremental polynomial method, are suggested here. Other mathematical techniques for calculating the crack growth rate are possible; the procedure used in calculating the growth rate shall be specified in the test report. The observed scatter in the fatigue crack growth rate data is influenced by the method of data reduction.

9.2.2 Secant method

Calculating the crack growth rate via the secant method entails computing the slope of a straight line connecting two adjacent data pairs of crack length and elapsed cycle count and represents an average velocity:

$$\frac{\mathrm{d}a\,(j)_{\text{avg.}}}{\mathrm{d}N} = \frac{(a_j - a_{j-1})}{(N_j - N_{j-1})} \tag{18}$$

for the incremental crack extension, $a_j - a_{j-1}$.

The stress intensity factor range is calculated using the average crack length over the increment of crack extension:

$$a\,(j)_{\text{avg.}} = \frac{(a_j + a_{j-1})}{2} \tag{19}$$

9.2.3 Incremental polynomial method

Calculating the crack growth rate by the incremental polynomial method (K-increasing only) requires fitting a polynomial to a segment of the data pairs: crack length, a_j, as a function of elapsed cycles, N_j. The data segment consists of an odd number of elements (3, 5 or 7), which are consecutive a_j versus N_j data pairs. The growth rate equals the slope of the polynomial, $\mathrm{d}a/\mathrm{d}N_j$, for the data segment's centremost element, e.g. for a data segment consisting of seven data pairs, the slope would be calculated as the derivative at the fourth element. The stress intensity factor range associated with the data segment is determined by using the fitted crack length of the centremost element of the data segment. For a data segment consisting of 3, 5 or 7 elements the fitted crack length corresponding to the second, third or fourth element, respectively, would be used in determining the stress intensity factor range for the data segment.

9.3 Determination of the fatigue crack growth threshold

The crack growth threshold, ΔK_{th}, generally refers to the asymptotic value of ΔK for which the corresponding $\mathrm{d}a/\mathrm{d}N$ approaches zero. It is commonly defined as being the value of ΔK corresponding to a crack growth rate equal to 10^{-8} mm/cycle [25], [48]. The fatigue crack growth rate corresponding to the threshold stress intensity factor range shall be reported. A common way to determine the threshold is to use a straight line fitted to a minimum of five, approximately equally spaced, $\log(\mathrm{d}a/\mathrm{d}N)$ versus $\log(\Delta K)$ data pairs between 10^{-7} mm/cycle and 10^{-8} mm/cycle; here, $\log(\Delta K)$ is the dependent variable of the best fit straight line. Using this linear regression technique, the value of ΔK is defined, by this test method, as the threshold stress intensity factor range, ΔK_{th}, at a fatigue crack growth rate equal to 10^{-8} mm/cycle.

In the case where data is generated within different fatigue crack growth rate ranges the above procedure may be used with the lowest decade of the $\mathrm{d}a/\mathrm{d}N$ test data.

10 Test report

10.1 General

The test report shall include a reference to this International Standard, i.e. ISO 12108, and the test date(s), plus the following information.

10.2 Material

a) standard alloy designation

b) thermal/mechanical conditioning

c) product form

d) chemical composition

e) 0,2 % proof stress used to evaluate specimen size criteria

f) ultimate tensile strength

g) modulus of elasticity (required when compliance crack length measurements are used)

10.3 Test specimen

a) specimen configuration

b) crack plane orientation (see Figure 11)

c) specimen location

d) width, W

e) thickness, B

f) notch height, h

g) stress intensity factor expression as a function of crack length and force

h) Specimen drawing and the reference source

10.4 Precracking terminal values

a) elapsed cycles at final stress intensity range

b) final crack extension

c) final crack length, a_{p}

d) final stress intensity factor range

e) final maximum stress intensity factor

f) maximum terminal stress intensity factor of the previous step

g) force ratio

h) cyclic waveform

10.5 Test conditions

a) testing machine force capacity

b) measurement cell force range

c) initial stress intensity factor range, ΔK_j

d) force ratio

e) forcing frequency

f) cyclic waveform

g) test procedure used (K-increasing or K-decreasing)

h) test environment

i) test temperature

j) laboratory relative humidity

k) measurement interval of a, Δa

l) crack curvature correction, a_{cor}

m) K-gradient if K-decreasing procedure is used

n) method of crack length measurement

10.6 Test analysis

a) analysis method for converting the crack length, a, and elapsed force cycles, N, data to crack growth rate, da/dN, i.e. the secant method, incremental polynomial method, etc.

b) remaining ligament size criteria used to assure predominant elastic loading in a non-standard test specimen configuration

c) report ΔK_{th} and the lowest decade of near threshold crack growth rate data used in its determination

d) exceptions to this test method

e) anomalies that could affect test results, e.g. test interruptions or changing the loading variables

10.7 Presentation of results

a) The results of the fatigue crack growth test shall be tabulated including: a_{fat}, a, N, ΔK and da/dN, as presented in Figure 15. Figure 15 can be expanded, as necessary, to include all measured crack lengths and forcing conditions. See Figure 17.

b) The results shall also be presented in a log-log plot with log (ΔK) plotted on the abscissa and log $(\mathrm{d}a/\mathrm{d}N)$ on the ordinate. For optimum data comparison it is recommended that the size of the log (ΔK) cycle be two to three times larger than that of the log $(\mathrm{d}a/\mathrm{d}N)$ cycle, as shown in Figure 16. For both the plot and the table, data violating the validity criteria shall be clearly identified. An example of the presentation of fatigue crack growth data is shown in Figures 17 and 18. When a negative force ratio ($R < 0$) is used, the method of calculating the stress intensity factor range, $\Delta K = (1 - R)\, K_{\text{max}}$ or $\Delta K = K_{\text{max}}$, shall be clearly identified on both the table and optional figure; also see 3.11 stress intensity factor range, ΔK.

FATIGUE CRACK GROWTH TEST RESULTS [REFERENCE ISO/TC 164/SC 5/WG 6]

Date.. Specimen I.D. Mark............................ Page 1 of ....

MATERIAL .... Product form ....
Thermal/mechanical conditioning ....
Chemical composition...

MECHANICAL PROPERTIES AT TEST TEMPERATURE
Ultimate tensile strength .... MPa 0,2 % Proof strength.. MPa
Modulus of elasticity....................................MPa

SPECIMEN
Dimensions: Thickness, B...................mm Width, W....................mm Machined notch length, a_n ....mm
Crack plane orientation .... Notch, h ....mm Specimen location ....
Stress intensity factor reference ....

PRECRACK TERMINAL VALUES
Final crack extension, Δa ....mm Preceding K_{max} (j–1) .... $MPa \cdot m^{1/2}$
Final crack length, a_p ....mm Force ratio ....
Final K_{max} (j) ....$MPa \cdot m^{1/2}$ Final load cycles ....
Final ΔK (j) ....$MPa \cdot m^{1/2}$ Cyclic waveform ....

TEST CONDITIONS
Test machine capacity .... Environment ....
Force transducer range...................................... Temperature ....°C
Measurement intertval of a, Δa ....mm Relative humidity ....%
Force frequency .... Hz Force ratio ....
Test procedure.. K-gradient .... mm^{-1}
Crack correction...mm Cyclic waveform ....
Initial ΔK_i ....$MPa \cdot m^{1/2}$ Initial force range .... kN
Crack measurement method ....

TEST ANALYSIS
Threshold stress intensity factor range, ΔK_{th}.. $MPa \cdot m^{1/2}$
Threshold crack growth rate decade ....mm/cycle
Analysis method...
Remaining size criteria Ref:...

EXCEPTIONS, ANOMALIES AND COMMENTS

Figure 15 — Test report

Page .... of ....

MATERIAL: .... SPECIMEN I.D. .... DATE ....

Measurement number	a_{fat} (mm)	N Cycles	a (mm)	$a(j)_{\text{avg.}}$ (mm)	$g(a/W)$ (mm)	ΔK (mm)	da/dN (mm/cycle)	

Figure 15 — Test report (*continued*)

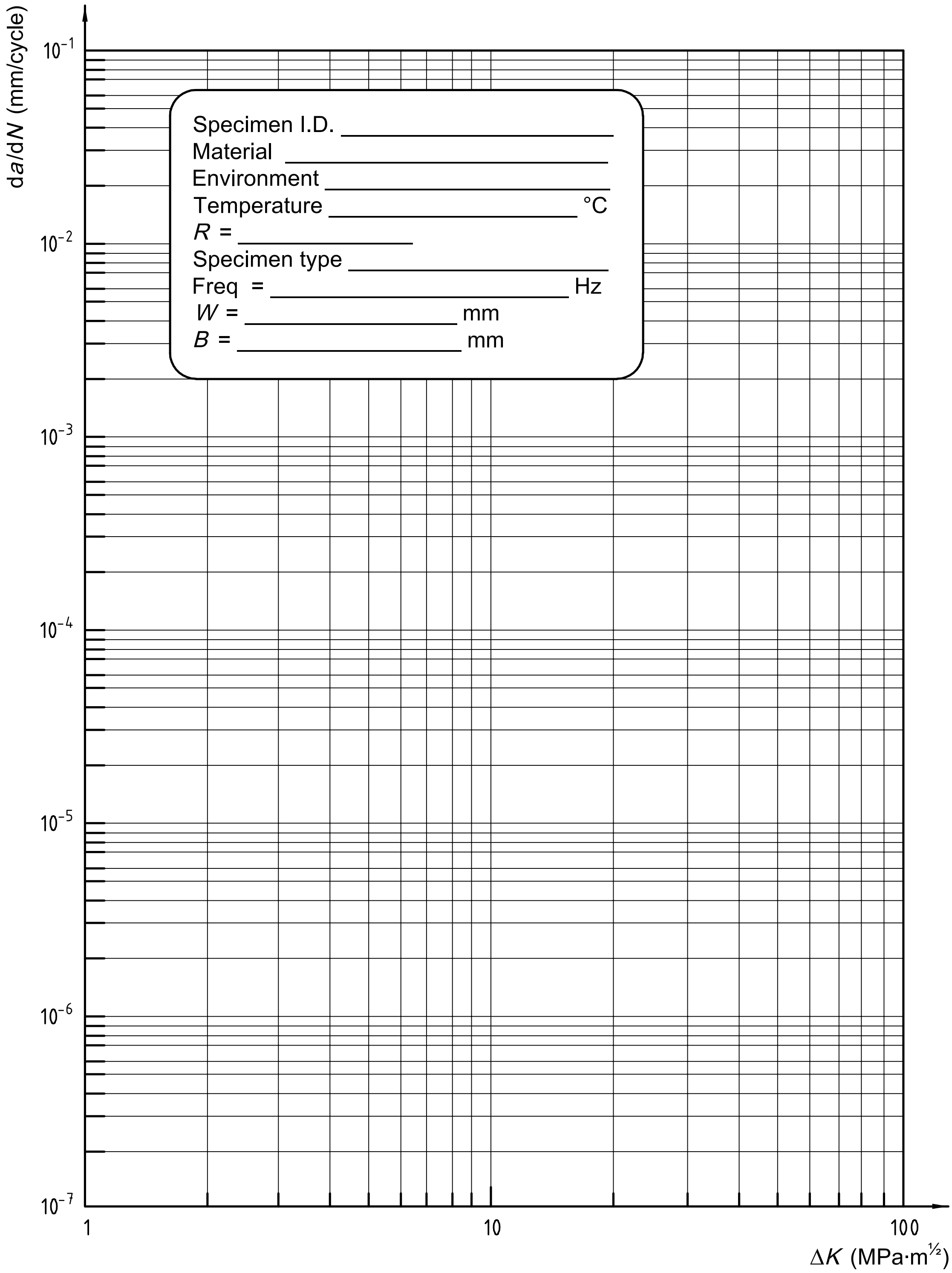


Figure 16 — Example axis for plotting $\log\left(\mathrm{d}a/\mathrm{d}N\right)$ **versus** $\log\left(\Delta K\right)$ **test data**

FATIGUE CRACK GROWTH TEST RESULTS [REFERENCE ISO/TC 164/SC 5/WG 6]

Date....................... 1/9/90 .... Specimen I.D. Mark.......... N410 .... Page 1 of .... 2...................

MATERIAL .... 16 MND 5......................... Product form .... forging ....
Thermal/mechanical conditioning ....
Chemical composition...

MECHANICAL PROPERTIES AT TEST TEMPERATURE
Ultimate tensile strength .... 611 MPa 0,2 % Proof strength.. 423 MPa
Modulus of elasticity.....................................MPa

SPECIMEN
Dimensions: Thickness, B.............25 mm Width, W..............50 mm Machined notch length, a_n ....10 mm
Crack plane orientation.......................Y-Z Notch, h ....1 mm Specimen location ....1/4 thickness
Stress intensity factor reference .... ISO/TC 164/SC 5/ WG 6 ....

PRECRACK TERMINAL VALUES
Final crack extension, Δa .... mm Preceding K_{max} (j–1) .... MPa·m$^{1/2}$
Final crack length, a_p ....12,50 mm Force ratio .... 0,1..
Final K_{max} (j) ....MPa·m$^{1/2}$ Final load cycles ....
Final ΔK (j) ....MPa·m$^{1/2}$ Cyclic waveform ....sinusoidal ....

TEST CONDITIONS
Test machine capacity .... Environment .... air..
Force transducer range...................................... Temperature ....300 °C
Measurement intertval of a, Δa ....1,2 mm Relative humidity ....%
Force frequency... 1 Hz Force ratio 0,1
Test procedure...............................K-increasing K-gradient .... mm^{-1}
Crack correction..0 mm Cyclic waveform ....sinusoidal ....
Initial ΔK_i .... 18,07 MPa·m$^{1/2}$ Initial force range .... 19,620 kN
Crack measurement method .... compliance ....

TEST ANALYSIS
Threshold stress intensity factor range, ΔK_{th}... MPa·m$^{1/2}$
Threshold crack growth rate decade ....mm/cycle
Analysis method.............secant...
Remaining size criteria Ref:....................................... ISO 12108

EXCEPTIONS, ANOMALIES AND COMMENTS

[1] Crack front curvature correction, a_{corr} = 0 mm

[2] Data violates remaining ligament size criteria

Figure 17 — Example of a test report

MATERIAL: .16 MND 5 .... SPECIMEN I.D. ...... N410 .... DATE .... 1/9/90....................

Measurement number	a_{fat} (mm)	N Cycles	a (mm)	$a(j)_{avg.}$ (mm)	$g(a/W)$ (mm)	ΔK (mm)	da/dN (mm/cycle)	
1	4,20	205 000	14,20					
2	5,60	237 080	15,60	14,90	5,59	19,62	4,36E-05	[1]
3	6,80	258 200	16,80	16,20	5,98	20,99	5,68E-05	
4	8,80	280 720	18,80	17,80	6,49	22,78	8,88E-05	
5	10,10	293 050	20,10	19,45	6,98	24,50	1,05E-04	
6	12,30	310 000	22,30	21,20	7,76	27,24	1,30E-04	
7	14,00	321 800	24,00	23,15	8,65	30,36	1,44E-04	
8	15,50	328 500	25,50	24,75	9,51	33,38	2,24E-04	
9	17,50	336 000	27,50	26,50	10,63	37,31	2,67E-04	
10	18,35	338 800	28,35	27,93	11,68	41,00	3,04E-04	
11	19,70	342 300	29,70	29,03	12,70	44,58	3,86E-04	
12	20,50	343 800	30,50	30,10	13,76	48,30	5,33E-04	
13	21,40	345 380	31,40	30,95	14,74	51,74	5,70E-04	
14	22,65	346 720	32,65	32,03	16,19	56,83	9,33E-04	[2]
15	23,80	347 730	33,80	33,23	18,07	63,43	1,14E-03	[2]
16	24,70	348 230	34,77	34,29	20,04	70,35	1,94E-03	[2]

Figure 17 — Example of a test report (*continued*)

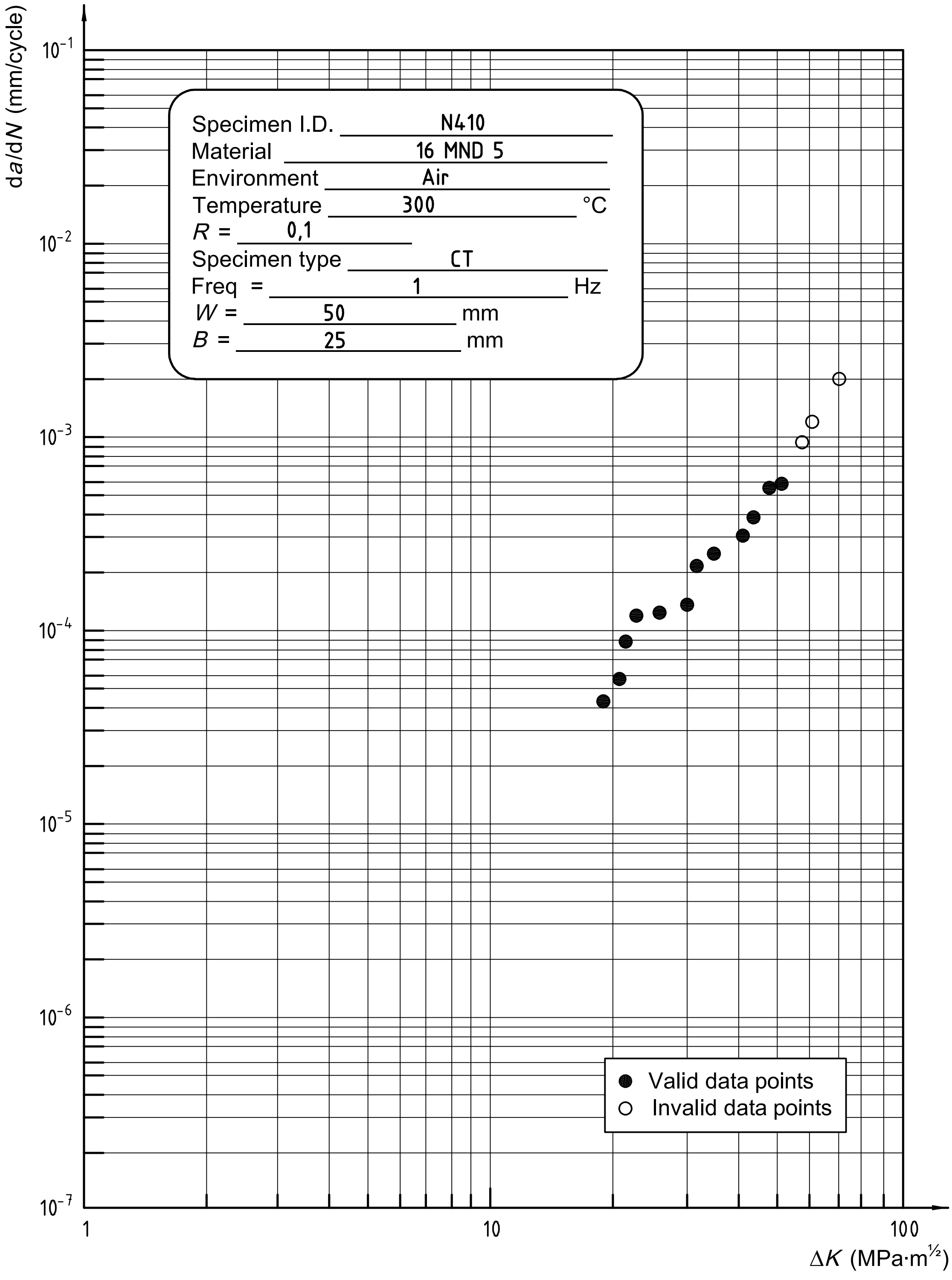


Figure 18 — Example plot of fatigue crack growth test result

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